

Claims

1. A combustion chamber (4) for a gas turbine (1), the combustion chamber wall (24) of which is provided on the inside with a lining formed from a plurality of heat shield elements (26), wherein the heat shield element or each heat shield element (26) forms with the combustion chamber wall (24) an inner space (27) which can be exposed to a cooling medium (K) and into which a flow element (49) is inserted for selective adjustment of a cooling medium stream.
2. The combustion chamber (4) as claimed in claim 1, wherein a flow channel (51) for cooling medium (K) is formed by the flow element (49), in which flow channel (51) the flow velocity (v_1) of the cooling medium stream is increased compared with the flow velocity (v_0) upstream of the flow element (49).
3. The combustion chamber (4) as claimed in claim 1 or 2, wherein a heat shield element (26) is assigned a respective flow element (49) for the purpose of cooling a thermally heavily loaded wall section (47A) of the heat shield element (26).
4. The combustion chamber (4) as claimed in claim 3, wherein the heat shield element (26) is a single-shell hollow vessel with a cavity (63) in which the flow element (49) is disposed.
5. The combustion chamber (4) as claimed in claim 3 or 4, wherein the heat shield element (26) has a surface region (57A, 57B) with a surface contour curved along a longitudinal axis (43) and a transverse axis (45).

6. The combustion chamber (4) as claimed in one of the claims 1 to 5, wherein the flow element (49) is mounted with a positive fit on the combustion chamber wall (24).

5 7. The combustion chamber (4) as claimed in one of the claims 1 to 6, wherein the flow element (49) is detachably connected to the combustion chamber wall (24).

8. The combustion chamber (4) as claimed in one of the claims
10 1 to 7, comprising a flow element (49) made of metal, in particular a metal sheet or a metal shaped part.

9. A gas turbine (1) having a combustion chamber (4) as claimed in one of the claims 1 to 8.